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THE UNITED STATES PATENT AND TRADEMARK OFFICE

lication of

Mark H. SHIPTON et al.

Group Art Unit: 1742

Application No.: 09/987,683

Filed: November 15, 2001

Docket No.: 111129

For:

NICKEL ALLOY COMPOSITION

CLAIM FOR PRIORITY

Director of the U.S. Patent and Trademark Office Washington, D.C. 20231

Sir:

The benefit of the filing date of the following prior foreign application filed in the following foreign country is hereby requested for the above-identified patent application and the priority provided in 35 U.S.C. §119 is hereby claimed:

British Patent Application No. 0028215.2 filed November 18, 2000.

In support of this claim, a certified copy of said original foreign application:

X	is filed herewith.
	was filed on in Parent Application No filed
	will be filed at a later date.

It is requested that the file of this application be marked to indicate that the requirements of 35 U.S.C. §119 have been fulfilled and that the Patent and Trademark Office kindly acknowledge receipt of this document.

Respectfully submitted,

James A. Oliff

Registration No. 27,075

Thomas J. Pardini Registration No. 30,411

JAO:TJP/cmm

Date: January 23, 2002

OLIFF & BERRIDGE, PLC P.O. Box 19928 Alexandria, Virginia 22320 Telephone: (703) 836-6400

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The Patent Office Concept House Cardiff Road Newport South Wales NP10 8QQ

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Signed

Dated

22 November 200

OLIFF & BERRIDGE, PLC
P.O. BOX 19928
ALEXANDRIA, VA 22320
(703) 836-6400
APPLICANT: Mark H. SHIPTON et al. APPLICATION NO.: 09/987,683
FILED: November 15, 2001
FOR: NICKEL ALLOY COMPOSITION
ATTORNEY DOCKET NO.: 111129

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Patents Act 1977 (Rule 16)

See note (d))

Patent Office

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Request for grant of a patent

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NEWPORT

The Patent Office

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PAT/VJB/2836 Your reference 0028215.2 2. Patent application number 18 NUT 200 (The Patent Office will fill in this part) ROLLS-ROYCE plc 3. Full name, address and postcode of the or of each applicant (underline all surnames) 65 BUCKINGHAM GATE LONDON SW1E 6AT **GREAT BRITAIN** Patents ADP number (if you know it) 570002 If the applicant is a corporate body, give the **GREAT BRITAIN** country/state of its incorporation NICKEL ALLOY COMPOSITION Title of the invention MR V J BIRD 5. Name of your agent (if you have one) INTELLECTUAL PROPERTY DEPT EW 1-5 "Address for service" in the United Kingdom ROLLS-ROYCE plc to which all correspondence should be sent PO BOX (including the postcode) 65779603 **FILTON BRISTOL BS34 7QE** Patents ADP number (if you know it) 6. If you are declaring priority from one or more Priority application number Date of filing Country earlier patent applications, give the country (if you know it) (day / month / year) and the date of filing of the or of each of these earlier applications and (if you know it) the or each application number 7. If this application is divided or otherwise Number of earlier application Date of filing derived from an earlier UK application, (day / month / year) give the number and the filing date of the earlier application YES 8. Is a statement of inventorship and of right to grant of a patent required in support of this request? (Answer 'Yes' if: a) any applicant named in part 3 is not an inventor, or b) there is an inventor who is not named as an applicant, or c) any named applicant is a corporate body.

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10. If you are also filing any of the following, state how many against each item.	
Priority documents	0
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Statement of inventorship and right to grant of a patent (Patents Form 7/77)	YES
Request for preliminary examination and search (Patents Form 9/77)	YES
Request for substantive examination (Patents Form 10/77)	NO
Any other documents (please specify)	NO
11.	I/We request the grant of a patent on the basis of this application.
	Signature Date (MR V J BIRD) 16 NOVEMBER 2000

Name and daytime telephone number of person to contact in the United Kingdom (MR V J BIRD) / / / MR V J BIRD 0117 979 7137

Warning

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NICKEL ALLOY COMPOSITION

The invention concerns a nickel alloy composition.

In particular, the invention concerns a nickel alloy composition for application to the tips of gas turbine blades to act as an oxidation and corrosion resistant barrier.

According to one aspect of the invention a nickel alloy has the following composition:

Element	Nominal Composition
	wt %
Co	4
Cr	4.5
W	2
Та	6
Re	4
Al	6
С	0.05
В	0.005
Hf	0.15
Si	0.1
La	0.003 - 0.005
Y	0.003 - 0.005
Ni	Remainder

A layer of nickel alloy having a composition substantially as referred to above may be applied to the radially outermost tip of a gas turbine blade. Such a layer is effective as a barrier to prevent oxidation of the hot nickel alloy and also to prevent corrosion damage in general. The layer is preferably applied by means of a laser cladding process or by weld deposition.

In a laser drilling process a laser beam is directed at a selected surface or surface region of a nickel alloy component and a finely divided powder having a composition of the subject nickel alloy is fed directly into the laser beam impingement zone. The

beam energy density is sufficient to melt the nickel alloy powder causing the powder particles to fuse to the surface of the component and to each other, thereby to form a homogeneous layer of relatively high density. Effectively the particles of nickel powder are sintered.

Alternatively, the nickel alloy of the intended barrier layer may be prepared in rod or wire form and fed into the plume or arc of a welding torch onto a blade tip surface. As before the powder particles are fused together and to the blade surface to form the required barrier layer.

The deposited nickel alloy layer provides a high integrity treatment for the tips of shroudless turbine blades in a gas turbine, in particular a gas turbine propulsion engine.

Furthermore the tip layer exhibits a useful level of rub tolerance and so may be used where a tip may be expected to occur from time to time. The alloy may also be used to repair cast turbine components such as turbine blades, turbine shroud segments and nozzle guide vanes.



CLAIMS

1 A nickel alloy having the following composition:

Element	Nominal Composition wt %
Co Cr W Ta Re Al C B Hf Si La Y	4 4.5 2 6 4 6 0.05 0.05 0.15 0.1 0.003 - 0.005 0.003 - 0.005 Remainder

- A nickel alloy as claimed in claim 1 in the form of a layer applied to the tip of a gas turbine blade.
- A nickel alloy as claimed in claim 2 wherein the layer is applied to the tip of a gas turbine blade by a laser cladding process.
- A nickel alloy as claimed in claim 2 wherein the layer is applied to the tip of a gas turbine blade by a weld deposition process.
- A nickel alloy substantially as hereinbefore described with reference to the disclosed composition.
- A gas turbine blade having a tip layer comprising nickel alloy as claimed in claim 5.

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